

Student Research Project / Master Thesis Flush Air Data System for Reusable Nanolaunchers

Currently, GAIA Aerospace is investigating the reusability of nanolauncher first stages. For both the ascent and the controlled re-entry of the first stage, an appropriate air data sensor system is required to detect the existing flow conditions at the vehicle. Over the years, clusters of pressure sensors, so-called Flush Air Data Systems (FADS), have proven to be a viable solution for hypersonic missiles and re-entry experiments. However, for these vehicles, re-entry is usually performed without retropropulsion, thus requiring only a single FADS at the nose of the system. In addition, a rocket hardly offers suitable positions for a FADS due to its cylindrical shape at the tail.

One possibility, however, is to accommodate a FADS in swept fins at the rear of the launcher. In this case, however, the shock fronts that form in the course of a re-entry must be taken into account in particular during data acquisition. Furthermore, the system also has to withstand the redirected exhaust jet of the engines during a re-entry burn. Therefore, within the scope of this work, a double-sided FADS is to be designed which meets the special requirements of a reusable nanolauncher.

The work is divided into the following steps:

- 1. Literature research on the aerodynamics of rockets and control surfaces up to Mach 10, pressure sensors, flush air data systems and CFD analyses.
- 2. Identification and definition of requirements for the Flush Air Data System
- 3. Definition of ascent and re-entry scenarios
- 4. Design of a FADS using COTS components in the form of a CAD model and a mathematical model for pressure data processing
- 5. Acquisition of FADS pressure data in the defined scenarios using CFD
- 6. Implementation, testing and optimization of the mathematical FADS model in a predefined flight simulation in a Matlab/Simulink environment
- 7. Critical analysis of the final system and presentation of further potential for optimization

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